Airplane Operating Manual Section 2 LIMITATIONS

This aircraft must be operated in accordance with the FAA-approved Operating Limitations (Op Limits). The Op Limits are printed on three typed pages. They form a part of the Airworthiness Certificate and as such must remain attached to the certificate.

INDICATED AIRSPEEDS				
KIAS	TERM	MARKING	DESCRIPTION	
52	V_{so}	Bottom of White Arc	Stall speed in landing configuration	
60	Vs	Bottom of Green Arc	Stall speed in cruise configuration	
87	V_{FE}	Top of White Arc	Flap extension speed	
125	V _A	Blue Line and Placard	Maximum speed for full control input at gross weight	
156	V_{NO}	Top of Green Arc	Normal operations in smooth air	
156-	V _{NO} -	Yellow Arc	Operation in smooth air with caution to avoid abrupt	
200	V_{NE}	Tellow Aic	control inputs	
200	V_{NE}	Red Line	Never exceed speed	

NOTE: Because of the high ratio of top speed to stall speed and Maneuvering speed, this aircraft is more susceptible to pilot induced overloads than other similar airplanes. THE PILOT CAN THEREFORE IMPOSE DESTRUCTIVE LOADS ON THE AIRFRAME ABOVE THE RELATIVELY LOW MANEUVERING SPEED. NOTE LIMITATIONS, EXERT CAUTION AND FLY ACCORDINGLY.

POWERPLANT LIIMITATIONS

Fuel Grade

Maximum Output	260 Horsepower
Maximum Speed	2700 RPM
Maximum Oil Temperature	235°F
Normal Oil Temp	140-220°F
Minimum Oil Pressure	20 psi
Maximum Oil Pressure	95 psi
Maximum Absolute Cylinder Head Temperature (CHT) Maximum Normal CHT Normal CHT	500°F 435°F 350°F -385°F
Maximum Fuel Pressure	45 psi
Minimum Fuel Pressure	14 psi

Leaded: 100LL as defined by ASTM D910
Unleaded: UL94 as defined by ASTM D7547
See Lycoming Service Instruction 1070AB

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ENGINE INSTRUMENT MARKINGS

Tachometer	Normal Operating Range Maximum	0-2700 RPM 2700 RPM
Oil Temperature	Normal Operating Range Maximum	140 to 220°F 245°F
Oil Pressure	Normal Operating Range Idle Range Minimum Maximum	45 to 95 psi 25 to 60 psi 20 psi 95 psi
Fuel Pressure	Normal Operating Range Minimum Maximum	14 to 45 psi 14 psi 45 psi
CHT Normal Maxin	435°F	

WEIGHT LIMITATIONS

Gross Weight 2700 pounds

Maximum baggage 150 pounds

CENTER OF GRAVITY LIMITS

The forward limit is 15% of wing chord or 8.4" aft of the wing leading edge. The aft limit is 30% of wing chord or 16.8" aft of the wing leading edge.

NOTE: Some RV-10s use an arbitrary datum somewhat forward of the spinner. *The datum for this aircraft is the wing leading edge.*

FLIGHT LOAD FACTORS

The structure has been designed to withstand aerodynamic loads of 3.8 G positive and 1.9 G negative. This is the maximum load the airframe structure is designed to withstand without sustaining damage. A significant safety factor is engineered into the limits. The calculated breaking strength is 5.7 G (plus safety factor) at which it will withstand the load for 3 seconds (assuming no previous airframe deterioration, fatigue, material flaws, or construction errors). Approaching this 5.7 G load could permanently weaken the structure even if failure does not occur.

TYPES OF APPROVED OPERATION

This aircraft is approved for Day and Night VFR and IFR operation in accordance with the aircraft Op Limits.

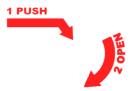
Airplane Operating Manual Section 2 LIMITATIONS

PLACARDS

Forward left fuselage near the external power port:



Outer left and right doors. Mirror and orient as necessary:



Left and right fuel cap:

AVGAS 30 GAL

Right instrument panel:

PASSENGER WARNING
THIS AIRCRAFT IS AMATEUR BUILT AND DOES NOT
COMPLY WITH FEDERAL SAFETY REGULATIONS
FOR STANDARD AIRCRAFT

Upper central instrument panel:





Knob face on right side of throttle quadrant:



RV-10

Airplane Operating Manual Section 2 LIMITATIONS

N921AC S/N 40394

Right instrument panel:

REDUCE V_A AT LOWER WEIGHTS:

2700 LBS 125 KIAS 2500 LBS 120 KIAS 2300 LBS 115 KIAS 2100 LBS 110 KIAS

Forward center console:

FIRE EXTINGUISHER INSIDE

Inside baggage door:

MAXIMUM BAGGAGE CAPACITY 150 LBS

Right rear window in 2-inch letters:



Left and right flap roots:

NO STEP

END OF SECTION